

To detect and prevent corrosion in the lower skins of the wing center section, which could reduce the integrity of the wing-to-fuselage fitting and consequently could lead to separation of the wing from the airplane, accomplish the following:

(a) For all airplanes: Within 4 weeks after July 3, 1971 (the effective date of AD 67-04-01, amendment 39-1234), remove the bottom wing center fairings having part numbers (P/N) 159W10400-121 and 159W10401-121, or use an FAA-approved equivalent method, to perform a visual inspection to detect corrosion of the wing planks under these fairings.

**Note 2:** Paragraph (a) of this AD merely restates the actions previously required by AD 67-04-01, amendment 39-1234. As allowed by the phrase, "unless accomplished previously," if those requirements of AD 67-04-01 have already been accomplished, this AD does not require that those actions be repeated.

**Note 3:** Care must be exercised when removing the fairings, since the attaching rivets go into the pressure vessel. Use caution not to enlarge rivet holes when removing rivets. When reinstalling the fairings, an adequate type fastener and sealant must be used.

**Note 4:** Grumman Service Newsletter, Volume 166, dated August-September 1966, pertains to this subject.

(b) For airplanes on which a protective paint system has not been installed in accordance with Grumman Gulfstream I Aircraft Service Change No. 190, dated June 28, 1971: Accomplish paragraphs (b)(1) and (b)(2) of this AD. As of the effective date of this AD, the inspections required by this paragraph shall be accomplished in accordance with Grumman Gulfstream I Aircraft Service Change No. 190, dated June 28, 1971.

**Note 5:** The repeated inspection referred to in this paragraph is the same inspection previously required by AD 67-04-01. Paragraph (b)(1) of this AD merely restates the requirement of AD 67-04-01 to repeat the inspection at intervals of 6 months. Paragraph (b)(2) permits the reinspection interval to be extended to 18 months once the specified protective paint system is installed.

(1) As a result of the inspection required by paragraph (a) of this AD:

(i) If no corrosion is detected, repeat the inspection thereafter at intervals not to exceed 6 months (26 weeks) until the actions specified in paragraph (b)(2) of this AD are accomplished.

(ii) If any corrosion is detected, prior to further flight, either repair the corroded part with an FAA-approved repair; or replace the corroded part with a new or serviceable part of the same part number; or replace the corroded part with a part approved by the FAA. Thereafter, continue to perform the inspection at intervals not to exceed 6 months (26 weeks) until paragraph (b)(2) of this AD is accomplished.

(2) Within 12 months after the effective date of this AD, install the protective paint system in accordance with Grumman Gulfstream I Aircraft Service Change No. 190, dated June 28, 1971. After installation, continue to perform the inspection required

by this paragraph at intervals not to exceed 18 months.

(c) For airplanes on which a protective paint system has been installed previously in accordance with Grumman Gulfstream I Aircraft Service Change

No. 190, dated June 28, 1971: Accomplish paragraphs (c)(1) and (c)(2) of this AD. As of the effective date of this AD, the inspections required by this paragraph shall be accomplished in accordance with Grumman Gulfstream I Aircraft Service Change No. 190, dated June 28, 1971.

**Note 6:** The repeated inspection referred to in this paragraph is the same inspection previously required by AD 67-04-01. Paragraph (c)(1) of this AD merely restates the requirement of AD 67-04-01 to repeat the inspection at intervals of 12 months. Paragraph (c)(2) permits the reinspection interval to be extended to 18 months.

(1) As a result of the inspection required by paragraph (a) of this AD:

(i) If no corrosion is detected, repeat the inspection thereafter at intervals not to exceed 12 months until paragraph (c)(2) of this AD is accomplished.

(ii) If any corrosion is detected, prior to further flight, either repair the corroded part with an FAA-approved repair; or replace the corroded part with a new or serviceable part of the same part number; or replace the corroded part with a part approved by the FAA. Thereafter, continue to perform the inspection at intervals not to exceed 12 months until paragraph (c)(2) of this AD is accomplished.

(2) Within 18 months since the last inspection accomplished in accordance with paragraph (c)(1) of this AD (i.e., the last inspection accomplished in accordance with AD 67-04-01), repeat the inspection specified in paragraph (c)(1) of this AD.

(i) If no corrosion is detected, repeat the inspection thereafter at intervals not to exceed 18 months.

(ii) If any corrosion is detected, prior to further flight, repair in accordance with the service change. After repair, continue to perform the inspection at intervals not to exceed 18 months.

(d)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Atlanta Aircraft Certification Office (ACO), FAA, Small Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Atlanta ACO.

(2) Alternative methods of compliance, approved previously in accordance with AD 67-04-01, amendment 39-1234, are approved as alternative methods of compliance with this AD.

**Note 7:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Atlanta ACO.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

(f) The actions shall be done in accordance with Grumman Gulfstream I Aircraft

Service Change No. 190, dated June 28, 1971. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Gulfstream Aerospace Corporation, Technical Operations Department, P.O. Box 2206, M/S D-10, Savannah, Georgia 31402-2206. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Atlanta Aircraft Certification Office, Small Airplane Directorate, Campus Building, 1701 Columbia Avenue, Suite 2-160, College Park, Georgia; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(g) This amendment becomes effective on September 5, 1997.

Issued in Renton, Washington, on July 25, 1997.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 97-20129 Filed 7-31-97; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 95-NM-228-AD; Amendment 39-10097; AD 97-16-06]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Model A300-600 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all Airbus Model A300-600 series airplanes, that requires an inspection to detect cracks of certain attachment holes; and installation of a new fastener and follow-on inspections or repair, if necessary. This amendment is prompted by reports of fatigue cracking found on the forward fitting of frame 47 at the level of the last fastener of the external angle fitting. The actions specified by this AD are intended to prevent such fatigue cracking, which could result in reduced structural integrity of the airframe.

**DATES:** Effective September 5, 1997.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 5, 1997.

**ADDRESSES:** The service information referenced in this AD may be obtained

from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Tim Backman, Aerospace Engineer, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (206) 227-2797; fax (206) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all Airbus Model A300-600 series airplanes was published in the **Federal Register** on April 15, 1996 (61 FR 16418). That action proposed to require a rotating probe inspection to detect cracks of the attachment holes H and I, and installation of a new fastener and follow-on inspections, if necessary.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the two comments received.

Both commenters support the proposed rule.

### Conclusion

After careful review of the available data, including the comments noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

### Cost Impact

The FAA estimates that 35 Airbus Model A300-600 series airplanes of U.S. registry will be affected by this AD, that it will take approximately 37 work hours per airplane to accomplish the required actions, and that the average labor rate is \$60 per work hour. The required kits for accomplishing the inspection will cost approximately \$75 per airplane. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$80,325, or \$2,295 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

### Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the rules docket. A copy of it may be obtained from the rules docket at the location provided under the caption **ADDRESSES**.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**97-16-06 Airbus Industrie:** Amendment 39-10097. Docket 95-NM-228-AD.

**Applicability:** All Model A300-600 series airplanes, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (f) of this AD. The

request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent fatigue cracking on the forward fitting of frame 47 at the level of the last fastener of the external angle fitting, which could result in reduced structural integrity of the airframe, accomplish the following:

(a) Perform a rotating probe inspection to detect cracks of the attachment holes H and I in accordance with Airbus Service Bulletin A300-57-6049, dated September 9, 1994, at the applicable time specified in paragraph (a)(1) or (a)(2) of this AD.

(1) For airplanes on which Airbus Modification 10454 (reference Airbus Service Bulletin A300-57-6050) has not been installed: Inspect prior to the accumulation of 13,800 total landings, or within 750 landings after the effective date of this AD.

(2) For airplanes on which Airbus Modification 10454 (reference Airbus Service Bulletin A300-57-6050) or Airbus Modification 10155 has been installed: Inspect prior to the accumulation of 18,700 total landings, or within 750 landings after the effective date of this AD.

(b) If no crack is found, prior to further flight, install a new fastener in accordance with Airbus Service Bulletin A300-57-6049, dated September 9, 1994. Repeat the rotating probe inspection thereafter at intervals not to exceed 5,600 landings.

(c) If any crack in hole I is found to be greater than 0.196 inches in length and/or depth, prior to further flight, repair it in accordance with a method approved by the

Manager, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate.

(d) If any crack in hole H is found to be greater than .062 inches in length, prior to further flight, repair it in accordance with a method approved by the Manager, Standardization Branch, ANM-113.

(e) If any crack in hole H or hole I is found to be less than or equal to the limits specified in paragraphs (c) and (d) of this AD, prior to further flight, repair it in accordance with Airbus Service Bulletin A300-57-6049, dated September 9, 1994.

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Standardization Branch, ANM-113.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Standardization Branch, ANM-113.

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to

a location where the requirements of this AD can be accomplished.

(h) The inspection, installation, and certain repair shall be done in accordance with Airbus Service Bulletin A300-57-6049, dated September 9, 1994. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(i) This amendment becomes effective on September 5, 1997.

Issued in Renton, Washington, on July 25, 1997.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 96-NM-130-AD; Amendment 39-10095; AD 97-16-04]

RIN 2120-AA64

#### **Airworthiness Directives; Saab Model SAAB SF340A and SAAB 340B Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment supersedes an existing airworthiness directive (AD), applicable to certain Saab Model SAAB SF340A and SAAB 340B series airplanes, that currently requires inspections to detect improper connections of the wire harness installation to the cartridges of the fire extinguishers in the engine nacelles, correction of any discrepancy, and modification of the wiring. This amendment adds a revised modification of that wiring, which, if accomplished, would terminate the inspections currently required by the existing AD. This amendment is prompted by reports indicating that, due to the removal of a certain clamp during maintenance, these fire extinguisher cartridges still could be connected incorrectly after the modification required by the existing AD has been accomplished. The actions specified by this AD are intended to prevent incorrect wiring of the cartridges, which would result in

inability of the fire extinguishers to jointly discharge extinguishing agent into a nacelle in the event of an engine fire.

**DATES:** Effective September 5, 1997.

The incorporation by reference of Saab Service Bulletin SAAB 340-26-015, revision 1, dated December 8, 1995, as listed in the regulations, is approved by the Director of the Federal Register as of September 5, 1997.

The incorporation by reference of Saab Service Bulletin SAAB 340-26-012, revision 1, dated October 5, 1993, as listed in the regulations, was approved previously by the Director of the Federal Register as of February 16, 1994 (59 FR 4575, February 1, 1994).

**ADDRESSES:** The service information referenced in this AD may be obtained from Saab Aircraft AB, SAAB Aircraft Product Support, S-581.88, Linköping, Sweden. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:**

Ruth Harder, Aerospace Engineer, Standardization Branch, ANM-113, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-1721; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) by superseding AD 94-03-06, amendment 39-8813 (59 FR 4575, February 1, 1994), which is applicable to certain Saab Model SAAB SF340A and SAAB 340B series airplanes, was published in the **Federal Register** on April 9, 1997 (62 FR 17127). The action proposed to continue to require repetitive inspections to detect improper connections of the wire harness installation to the cartridges of the fire extinguishers in the engine nacelles, and correction of any discrepancies. The action also proposed to require a revised modification of the wiring to one of the electrical connectors if it exceeds a certain length. Accomplishment of the modification would terminate the repetitive inspections of the wiring currently required by AD 94-03-06.

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the single comment received.

The commenter supports the proposed rule.

### Conclusion

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

### Cost Impact

There are approximately 235 Saab Model SAAB SF340A and SAAB 340B series airplanes of U.S. registry that will be affected by this AD.

The actions that are currently required by AD 94-03-06 take approximately 6 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the previously required actions on U.S. operators is estimated to be \$77,760, or \$360 per airplane. (At the time AD 94-03-06 went into effect, it was estimated that 216 airplanes would be affected.)

The new actions that are required by this new AD will take approximately 6 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Required parts will be provided at no cost to operators. Based on these figures, the cost impact of the new requirements of this AD on U.S. operators is estimated to be \$84,600, or \$360 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

### Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy