

was published in the **Federal Register** on January 12, 2000 (65 FR 1831). That action proposed to establish life limits for stage 1 HPT and stage 1 LPT disks operated under the new flight plan profiles, C and D; require the removal from service of stage 1 HPT and stage 1 LPT disks prior to reaching new, lower cyclic life limits; and replace those disks with serviceable parts in accordance with R-R Service Bulletin TAY-72-1479, dated July 20, 1999.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received on the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### Economic Analysis

There are approximately 242 engines of the affected design in the worldwide fleet. The FAA estimates that three engines installed on aircraft of U.S. registry will be affected by this AD, and that the prorated life reduction would cost \$26,658 per engine. Based on these figures, the total cost impact of the proposed AD on US operators is estimated to be \$79,974.

#### Regulatory Impact

This rule does not have federalism implications, as defined in Executive Order (EO) No. 13132, because it does not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Accordingly, the FAA has not consulted with state authorities prior to publication of this rule.

For the reasons discussed above, I certify that this action (1) Is not a "significant regulatory action" under EO No. 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2000-08-01 Rolls-Royce plc:** Amendment 39-11687. Docket 99-NE-61-AD.

*Applicability:* Rolls-Royce plc (R-R) Tay 650-15 series turbofan engines, with stage 1 high pressure turbine (HPT) disks, part numbers (P/Ns) JR32013 and JR33838, and stage 1 low pressure turbine (LPT) disks, P/N JR32318A. These engines are installed on but not limited to Fokker F.28 Mark 0100 (F100) series aircraft.

**Note 1:** This airworthiness directive (AD) applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

*Compliance:* Required as indicated, unless accomplished previously.

To prevent crack initiation and propagation leading to turbine disk failure, which could result in an uncontained engine failure and damage to the aircraft, accomplish the following:

#### Flight Plan Profile C

(a) Remove from service stage 1 HPT disks, P/Ns JR32013 and JR33838, and stage 1 LPT disks, P/N JR32318A, operated under flight plan profile C, as defined in the R-R Tay Engine Manual, 70-01-10, pages 1-10, prior to accumulating 18,000 cycles-since-new (CSN), and replace with serviceable parts.

#### Flight Plan Profile D

(b) Remove from service stage 1 HPT disks, P/Ns JR32013 and JR33838, and stage 1 LPT disks, P/N JR32318A, operated under flight plan profile D, as defined in the R-R Tay Engine Manual, 70-01-10, pages 1-10, prior to accumulating 14,250 CSN, and replace with serviceable parts.

#### Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office (ECO). Operators shall submit their request through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, ECO.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the ECO.

#### Special Flight Permits

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the aircraft to a location where the requirements of this AD can be accomplished.

#### Incorporation by Reference Material

(e) The actions of this AD shall be done in accordance with R-R Service Bulletin TAY-72-1479, dated July 20, 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Rolls-Royce plc, P.O. Box 31, Derby, DE24 8BJ, UK, telephone 011-44-1332-242424. Copies may be inspected at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

#### Effective Date of This AD

(f) This amendment becomes effective on June 19, 2000.

Issued in Burlington, Massachusetts, on April 7, 2000.

**David A. Downey,**

*Assistant Manager, Engine and Propeller Directorate, Aircraft Certification Service.*

[FR Doc. 00-9358 Filed 4-17-00; 8:45 am]

**BILLING CODE 4910-13-U**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 99-NM-304-AD; Amendment 39-11682; AD 2000-07-26]

RIN 2120-AA64

#### Airworthiness Directives; Airbus Model A300 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A300 series airplanes, that requires a

one-time detailed visual inspection to detect corrosion on the outer surface of the fuselage skin panel; application of corrosion preventive protection; and corrective action, if necessary. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to detect and correct corrosion of the fuselage skin panel, which could result in cracking and consequent reduced structural integrity of the airplane.

**DATES:** Effective May 23, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the **Federal Register** as of May 23, 2000.

**ADDRESSES:** The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A300 series airplanes was published in the **Federal Register** on January 3, 2000 (65 FR 91). That action proposed to require a one-time detailed visual inspection to detect corrosion on the outer surface of the fuselage skin panel; application of corrosion preventive protection; and corrective action, if necessary.

#### Comment Received

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the single comment received.

#### Request to Reference Latest Service Bulletin Revision

One commenter, the manufacturer, requests that the proposed AD be revised to refer to Airbus Service Bulletin A300-53-0328, Revision 01, including Appendix 01, dated March

15, 2000, for accomplishment of the inspection. The original issue of the service bulletin, dated March 5, 1999, was referenced in the proposed AD as the appropriate source of service information. The commenter notes that the original issue of the service bulletin references a 30-month interval rather than the correct 5-year interval for certain follow-on repetitive inspections that are covered by the Corrosion Prevention Control Program (CPCP). The commenter suggests that referencing Revision 01 of the service bulletin, in which the correct interval is specified, will avoid confusion on the part of operators.

The FAA concurs. The FAA has reviewed the procedures described in Airbus Service Bulletin A300-53-0328, Revision 01, including Appendix 01, dated March 15, 2000, and has determined that they are equivalent to those described in the original issue of the service bulletin, except for certain cleaning procedures. The final rule has been revised to refer to Revision 01 of the service bulletin as the appropriate source of service information. However, a "NOTE" has been included in the final rule to provide credit for previous accomplishment of the actions required by this AD in accordance with the original issue of the service bulletin.

#### Conclusion

After careful review of the available data, including the comment noted above, the FAA has determined that air safety and the public interest require the adoption of the rule with the change described previously. The FAA has determined that this change will neither increase the economic burden on any operator nor increase the scope of the AD.

#### Cost Impact

The FAA estimates that 3 airplanes of U.S. registry will be affected by this AD, that it will take approximately 4 or 22 work hours per airplane, depending on the airplane configuration, to accomplish the required inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$240 or \$1,320 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

#### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) Is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2000-07-26 Airbus Industrie:** Amendment 39-11682. Docket 99-NM-304-AD.

**Applicability:** Model A300 series airplanes, certificated in any category; except those on which Airbus Modification 04201 has been accomplished.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD.

The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To detect and correct corrosion of the fuselage skin panel, which could result in cracking and consequent reduced structural integrity of the airplane, accomplish the following:

#### Inspection

(a) Perform a one-time detailed visual inspection of the outer surface of the fuselage skin panel between fuselage frames FR39 and FR40, and between stringers 27 and 33, for corrosion; in accordance with Airbus Service Bulletin A300-53-0328, Revision 01, including Appendix 01, both dated March 15, 2000. Perform the inspection at the applicable time specified in paragraph (a)(1), (a)(2), or (a)(3) of this AD. If any corrosion is found, prior to further flight, repair (*i.e.*, rework corroded areas, or repair or replace panels, as applicable) in accordance with the service bulletin, except as provided by paragraph (b) of this AD. Temporary repairs must be replaced with permanent repairs prior to accumulation of the life limits specified in the service bulletin.

(1) For airplanes for which the date of manufacture was less than 15 years before the effective date of this AD: Inspect within 18 months after the effective date of this AD.

(2) For airplanes for which the date of manufacture was at least 15 but less than 20 years before the effective date of this AD: Inspect within 12 months after the effective date of this AD.

(3) For airplanes for which the date of manufacture was 20 or more years before the effective date of this AD: Inspect within 6 months after the effective date of this AD.

**Note 2:** For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

(b) Where Airbus Service Bulletin A300-53-0328, Revision 01, dated March 15, 2000, specifies that Airbus may be contacted for a repair, prior to further flight, replace the skin panel with a new or serviceable skin panel in accordance with the service bulletin.

**Note 3:** Accomplishment of the actions required by this AD in accordance with Airbus Service Bulletin A300-53-0328, dated March 5, 1999, prior to the effective date of this AD, is acceptable for compliance with the requirements of this AD.

#### Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be

used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 4:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

#### Special Flight Permits

(d) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

#### Incorporation by Reference

(e) The actions shall be done in accordance with Airbus Service Bulletin A300-53-0328, Revision 01, including Appendix 01, dated March 15, 2000. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 5:** The subject of this AD is addressed in French airworthiness directive 1999-209-281(B), dated May 19, 1999.

(f) This amendment becomes effective on May 23, 2000.

Issued in Renton, Washington, on April 6, 2000.

**Donald L. Riggin,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 00-9112 Filed 4-17-00; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

**[Docket No. 2000-NM-83-AD; Amendment 39-11683; AD 2000-07-27]**

**RIN 2120-AA64**

#### **Airworthiness Directives; Various Transport Category Airplanes Equipped With Certain Honeywell Air Data Inertial Reference Units**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) that is applicable to various transport category airplanes equipped with certain Honeywell air data inertial reference units (ADIRU). This action requires inspection of a failed ADIRU to determine its modification status, and replacement of an unmodified failed ADIRU with a serviceable ADIRU. This action also provides for optional terminating action for the requirements of the AD. This amendment is prompted by reports of dual critical failures of inertial reference units on ADIRU's during flight. The actions specified in this AD are intended to prevent loss of the main sources of attitude data, consequent high pilot workload, and a significant increase in the likelihood of pilot error.

**DATES:** Effective May 3, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of May 3, 2000.

Comments for inclusion in the Rules Docket must be received on or before June 19, 2000.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2000-NM-83-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

The service information referenced in this AD may be obtained from Honeywell, Publications, P.O. Box 21111, Mail Stop DV-10, Phoenix, Arizona 85036. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Chicago Aircraft Certification Office, 2350 East Devon Avenue, Room 323, Des Plaines, Illinois; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

#### **FOR FURTHER INFORMATION CONTACT:**

Wess Rouse, Aerospace Engineer, Systems and Flight Test Branch, ACE-117C, FAA, Chicago Aircraft Certification Office, 2350 East Devon Avenue, Room 323, Des Plaines, Illinois 60018; telephone (847) 294-8113; fax (847) 294-7834.

**SUPPLEMENTARY INFORMATION:** The FAA has recently received three reports of dual inertial reference (IR) critical faults of the air data inertial reference system comprising two or more air data inertial reference units (ADIRU) on transport category airplanes during flight. Three days prior to one of the dual IR critical fault incidents, one of those ADIRU's