

within 30 days after the effective date of this AD; and, if any part is incorrect, limit the airplane ceiling to 31,000 feet until the incorrect part is replaced, as specified by paragraph (b) of this AD.

(2) For airplanes pre-Modification HCM50258A: Identify the part numbers within 6 months after the effective date of this AD.

#### Corrective Action

(b) For any incorrect part identified in accordance with paragraph (a) of this AD: Within 500 flight cycles thereafter, replace it with a new, correct part, in accordance with BAe Systems (Operations) Limited Inspection Service Bulletin ISB.21-148, Revision 1, dated February 6, 2001. Prior to further flight thereafter, perform a structural inspection and accomplish applicable corrective actions, in accordance with a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the Civil Aviation Authority (CAA) (or its delegated agent).

**Note 2:** Accomplishment of the actions specified in this AD in accordance with BAe Systems (Operations) Limited Inspection Service Bulletin ISB.21-148, dated November 17, 2000, is also acceptable for compliance with the requirements of this AD.

#### Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

#### Special Flight Permits

(d) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

#### Incorporation by Reference

(e) Except as required by paragraph (b) of this AD: The actions shall be done in accordance with BAe Systems (Operations) Limited Inspection Service Bulletin ISB.21-148, Revision 1, dated February 6, 2001. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from British Aerospace Regional Aircraft American Support, 13850 Mclearen Road, Herndon, Virginia 20171. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 4:** The subject of this AD is addressed in British airworthiness directive 003-11-2000.

#### Effective Date

(f) This amendment becomes effective on September 10, 2001.

Issued in Renton, Washington, on July 25, 2001.

**Vi L. Lipski,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 01-19249 Filed 8-3-01; 8:45 am]

**BILLING CODE 4910-13-U**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2000-NM-383-AD; Amendment 39-12357; AD 2001-15-22]

**RIN 2120-AA64**

#### Airworthiness Directives; Airbus Model A319, A320, and A321 Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A319, A320, and A321 series airplanes, that requires modifications of route segregation between the low voltage wire bundles of the fuel quantity indicating system and the high voltage wire bundles of the ground power control unit. This amendment is prompted by mandatory continuing airworthiness information from a civil airworthiness authority. The actions specified by this AD are intended to prevent injection of 115 volt alternating current (VAC) into 28 volt direct current (VDC) wire bundles, which could result in high voltage conditions within the fuel tank and the potential for damage to equipment, electrical arcing, and fuel vapor ignition on the ground. This action is intended to address the identified unsafe condition.

**DATES:** Effective September 10, 2001.

The incorporation by reference of Airbus Service Bulletin A320-92-1007, Revision 02, dated August 4, 2000, as listed in the regulations, is approved by the Director of the Federal Register as of September 10, 2001.

**ADDRESSES:** The service information referenced in this AD may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation

Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Tim Dulin, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2144; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A319, A320, and A321 series airplanes was published in the **Federal Register** on May 15, 2001 (66 FR 26815). That action proposed to require modifications of route segregation between the low voltage wire bundles of the fuel quantity indicating system and the high voltage wire bundles of the ground power control unit.

#### Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public. The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

#### Cost Impact

The FAA estimates that 291 airplanes of U.S. registry will be affected by this AD, that it will take approximately 24 to 42 work hours per airplane to accomplish the required modifications, depending on the wiring configuration of the airplane, and that the average labor rate is \$60 per work hour. Required parts will cost approximately \$1,300 per airplane. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be between \$797,340 and \$1,111,620, or between \$2,740 and \$3,820 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up,

planning time, or time necessitated by other administrative actions.

### Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

### Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2001-15-22 Airbus Industrie:** Amendment 39-12357. Docket 2000-NM-383-AD.

**Applicability:** Model A319, A320, and A321 series airplanes; certificated in any category; except those on which Airbus Industrie Modification 28289 has been installed.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance

of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent injection of 115 volt alternating current (VAC) into 28 volt direct current (VDC) wire bundles, which could result in high voltage conditions within the fuel tank and the potential for damage to equipment, electrical arcing, and fuel vapor ignition on the ground, accomplish the following:

### Modification

(a) Within 4 years after the effective date of this AD, install additional protective conduits and new supports to ensure physical route segregation between the low voltage wire bundles of the fuel quantity indicating system (FQIS) and the high voltage wire bundles of the ground power control unit (GPCU), in accordance with Airbus Service Bulletin A320-92-1007, Revision 02, dated August 4, 2000.

**Note 2:** Modifications accomplished prior to the effective date of this AD in accordance with Airbus Service Bulletin A320-92-1007, dated January 12, 2000; or Airbus Service Bulletin A320-92-1007, Revision 01, dated June 29, 2000; are considered acceptable for compliance with the applicable actions specified in this amendment.

### Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send them to the Manager, International Branch, ANM-116.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

### Special Flight Permits

(c) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

### Incorporation by Reference

(d) The modification shall be done in accordance with Airbus Service Bulletin A320-92-1007, Revision 02, dated August 4, 2000. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France.

Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 4:** The subject of this AD is addressed in French airworthiness directive 2000-407-150(B), dated September 20, 2000.

### Effective Date

(e) This amendment becomes effective on September 10, 2001.

Issued in Renton, Washington, on July 25, 2001.

**Vi L. Lipski,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 01-19250 Filed 8-3-01; 8:45 am]

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 2000-NM-274-AD; Amendment 39-12360; AD 2001-15-25]

RIN 2120-AA64

### Airworthiness Directives; Raytheon Model Hawker 800XP Series Airplanes and Model Hawker 800 (U-125A Military) Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to certain Raytheon Model Hawker 800XP series airplanes and certain Model Hawker 800 (U-125A military) airplanes, that requires a one-time inspection of an attachment bolt in the main landing gear (MLG) door system to determine whether the bolt's protruding threads have been peened; and corrective action, if necessary. The actions specified by this AD are intended to prevent the disconnection of the retaining hook (which holds the MLG door up and locked) from its means of actuation, which could result in a gear-up landing and possible injury to passengers and crew. This action is intended to address the identified unsafe condition.

**DATES:** Effective September 10, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 10, 2001.

**ADDRESSES:** The service information referenced in this AD may be obtained from Raytheon Aircraft Company,