

(RFM) or make the following pen and ink addition to the RFM Emergency Procedure for fuel filter clogged caution light illumination: "If both fuel filter clogged caution lights illuminate, land as soon as possible."

(c) If both filter clogged caution lights illuminate in flight, after landing, either:

(1) Accomplish the requirements of paragraphs (a)(1) through (a)(4) of this AD before further flight, or,

(2) Replace both fuel filter elements with airworthy fuel filter elements and conduct a one-time direct flight to a location where the requirements of paragraphs (a)(1) through (a)(4) of this AD can be accomplished before further flight.

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Regulations Group, Rotorcraft Directorate, FAA.

Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, Regulations Group.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Manager, Regulations Group.

(e) Special flight permits will not be issued.

(f) The inspections shall be done in accordance with the Accomplishment Instructions, paragraphs 2.B.1, 2.B.2., and 2.B.4. of Eurocopter France Alert Service Bulletin No. 01.00.56 Revision 1, dated March 15, 2001. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from American Eurocopter Corporation, Technical Support, 2701 Forum Drive, Grand Prairie, Texas 75053-4005. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 4: The subject of this AD is addressed in Direction Generale De L'Aviation Civile (France) AD's 1998-318-071(A)R6 and 1998-319-012(A)R6, both dated April 18, 2001.

Issued in Fort Worth, Texas, on August 17, 2001.

Eric Bries,

Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 01-21753 Filed 8-28-01; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-112-AD]

RIN 2120-AA64

Airworthiness Directives; Bombardier Model DHC-8-100, -200, and -300 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Bombardier Model DHC-8-100, -200, and -300 series airplanes. This proposal would require repetitive inspections of the rudder pedal adjustment fittings for cracks and replacement of cracked fittings with new fittings. The proposal would also provide an optional terminating action. This action is necessary to detect and correct cracking of the rudder pedal adjustment fittings, which could lead to deformation of the fittings, resulting in jammed rudder pedals and loss of rudder control, with consequent reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by September 28, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket Number 2001-NM-112-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmccomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2001-NM-112-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in the proposed rule may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. This information may be examined at the FAA, Transport Airplane

Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York.

FOR FURTHER INFORMATION CONTACT: Dan Parrillo, Aerospace Engineer, ANE-172, FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York 11581; telephone (516) 256-7505; fax (516) 568-2716.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (*e.g.*, reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2001-NM-112-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket

Number 2001–NM–112–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056.

Discussion

Transport Canada Civil Aviation (TCCA), which is the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on certain Bombardier Model DHC–8 –100, –200, and –300 series airplanes. The TCCA advises that three incidents have been reported of stiff rudder pedal operation. These incidents were due to cracking of the rudder pedal adjustment fittings. If not corrected, cracking of the rudder pedal adjustment fittings can lead to deformation of the fittings, resulting in jammed rudder pedals and loss of rudder control, with consequent reduced controllability of the airplane.

Explanation of Relevant Service Information

Bombardier, Inc., has issued Alert Service Bulletin A8–27–91, dated September 12, 2000, and Revision A, dated November 23, 2000, which describe procedures for a detailed visual inspection of the rudder pedal adjustment fittings for cracks and replacement of cracked fittings with new aluminum or steel fittings. New aluminum fittings would need to be inspected within 5,000 flight hours after installation and thereafter at intervals not to exceed 1,000 flight hours. Replacement with steel fittings would constitute terminating action for the purposes of this AD. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition.

The TCCA classified this service bulletin as mandatory and issued Canadian airworthiness directive CF–2001–04, dated January 25, 2001, in order to assure the continued airworthiness of these airplanes in Canada. That airworthiness directive specifies that, if there is any instance of stiff operation or jamming of the rudder pedals during flight, the rudder pedal adjustment fittings must be inspected in accordance with the Bombardier service bulletin prior to further flight.

FAA's Conclusions

This airplane model is manufactured in Canada and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the TCCA has kept the FAA informed of the situation described above. The FAA has examined the findings of the TCCA,

reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously.

Cost Impact

The FAA estimates that 188 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 1 work hour per airplane to accomplish the proposed inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$11,280, or \$60 per airplane, per inspection cycle.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this proposed AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this

action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Bombardier, Inc. (Formerly de Havilland, Inc.) Docket 2001–NM–112–AD.

Applicability: Model DHC–8–100, –200, and –300 series airplanes, serial numbers 003 to 563 inclusive, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect and correct cracking of the rudder pedal adjustment fittings, which could lead to deformation of the fittings, resulting in jammed rudder pedals and loss of rudder control, with consequent reduced controllability of the airplane, accomplish the following:

Inspection

(a) Inspect the rudder pedal adjustment fittings having part number (P/N) 82710038–101 for cracks, in accordance with Bombardier Service Bulletin A8–27–91, dated September 12, 2000, or Revision A, dated November 23, 2000, at the times specified in paragraphs (a)(1) and (a)(2) of this AD.

(1) Within 5,000 flight hours since the date of manufacture of the airplane or 500 flight hours after the effective date of this AD, whichever occurs later, and

(2) Prior to further flight, whenever an instance of stiff operation or jamming of the rudder pedals occurs during flight.

(b) If no cracks are detected: Repeat the inspection of the rudder pedal adjustment fittings having P/N 82710038-101, in accordance with Bombardier Service Bulletin A8-27-91, dated September 12, 2000, or Revision A, dated November 23, 2000, at intervals not to exceed 1,000 flight hours, until accomplishment of paragraph (d) of this AD.

(c) If cracks are detected: Prior to further flight, replace the rudder pedal adjustment fittings having P/N 82710038-101 with new aluminum fittings having the same P/N (82710038-101), or with steel fittings having P/N 82710080-101, in accordance with Bombardier Service Bulletin A8-27-91, dated September 12, 2000, or Revision A, dated November 23, 2000.

Terminating Action

(d) Replacement of the rudder pedal adjustment fittings having P/N 82710038-101, with steel rudder pedal adjustment fittings having P/N 82710080-101, constitutes terminating action for the repetitive inspections required by paragraphs (a) and (b) of this AD.

Alternative Methods of Compliance

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, New York Aircraft Certification Office, FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, New York ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the New York ACO.

Special Flight Permits

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Note 3: The subject of this AD is addressed in Canadian airworthiness directive CF-2001-04, dated January 25, 2001.

Issued in Renton, Washington, on August 22, 2001.

Ali Bahrami,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 01-21752 Filed 8-28-01; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-130-AD]

RIN 2120-AA64

Airworthiness Directives; McDonnell Douglas Model MD-90-30 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain McDonnell Douglas Model MD-90-30 series airplanes. This proposal would require installation of two arcing protection brackets below and behind the circuit breakers located in the generator control rack in the electrical/electronics compartment. This action is necessary to prevent arcing between circuit breaker terminals and adjacent equipment and structure located in the generator control rack in the electrical/electronics compartment, which, if not corrected, could result in possible electrical shock to maintenance personnel during maintenance operations. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by October 15, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2001-NM-130-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2001-NM-130-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in the proposed rule may be obtained from Boeing Commercial Aircraft Group, Long Beach Division, 3855 Lakewood Boulevard, Long Beach, California 90846, Attention: Data and Service Management, Dept. C1-L5A (D800-

0024). This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California.

FOR FURTHER INFORMATION CONTACT:

George Y. Mabuni, Aerospace Engineer, Systems and Equipment Branch, ANM-130L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5341; fax (562) 627-5210.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

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- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (e.g., reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2001-NM-130-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the