

3. Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and

4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

Airbus Helicopters: Docket No. FAA–2015–6033; Directorate Identifier 2015–SW–019–AD.

(a) Applicability

This AD applies to Airbus Helicopters Model AS 365 N3 helicopters, certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as missing or incorrectly located information for exiting a helicopter. This condition could result in failure to jettison cabin doors during an emergency, resulting in death or injury of helicopter occupants.

(c) Comments Due Date

We must receive comments by May 10, 2016.

(d) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(e) Required Actions

Within 50 hours time-in-service:

(1) Inspect the cabin and cockpit for labels, placards, and markings that provide jettison procedure instructions for cabin doors.

(2) For the left and right side, remove any existing label, placard, and marking and install placards in accordance with the Accomplishment Instructions, paragraph 3.B.2 and Figures 1 through 6, of Airbus Helicopters Alert Service Bulletin No.

AS365–11.00.02, Revision 2, dated April 23, 2015.

(f) Credit for Previously Completed Actions

Actions accomplished before the effective date of this AD in accordance with Airbus Helicopters Modification (MOD) 0711B68 for helicopters without external life rafts or MOD 0711B68 and MOD 0711B67 for helicopters with external life rafts are considered acceptable for compliance with this AD.

(g) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Group, FAA, may approve AMOCs for this AD. Send your proposal to: David Hatfield, Aviation Safety Engineer, Safety Management Group, Rotorcraft Directorate, FAA, 10101 Hillwood Pkwy, Fort Worth, TX 76177; telephone (817) 222–5110; email 9–ASW–FTW–AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office before operating any aircraft complying with this AD through an AMOC.

(h) Additional Information

The subject of this AD is addressed in the European Aviation Safety Agency (EASA) AD No. 2015–0068–E, dated April 29, 2015. You may view the EASA AD on the Internet at <http://www.regulations.gov> by searching for and locating it in Docket No. FAA–2015–6033.

(i) Subject

Joint Aircraft Service Component (JASC) Code: 1100, Placards and Markings.

Issued in Fort Worth, Texas, on February 29, 2016.

Scott A. Horn,

Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 2016–05369 Filed 3–10–16; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2016–4278; Directorate Identifier 2012–SW–022–AD]

RIN 2120–AA64

Airworthiness Directives; AgustaWestland S.p.A. (Agusta) Helicopters

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to adopt a new airworthiness directive (AD) for certain

Agusta Model AB139 and AW139 helicopters. This proposed AD would require performing operational checks of both hydraulic systems. This proposed AD is prompted by an assessment of the hydraulic systems of the helicopter following an accident. The proposed actions are intended to prevent loss of hydraulic power to the flight controls and subsequent loss of control of the helicopter.

DATES: We must receive comments on this proposed AD by May 10, 2016.

ADDRESSES: You may send comments by any of the following methods:

- **Federal eRulemaking Docket:** Go to <http://www.regulations.gov>. Follow the online instructions for sending your comments electronically.

- **Fax:** 202–493–2251.

- **Mail:** Send comments to the U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590–0001.

- **Hand Delivery:** Deliver to the “Mail” address between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket number FAA–2016–4278 or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the European Aviation Safety Agency (EASA) AD, the economic evaluation, any comments received, and other information. The street address for the Docket Operations Office (telephone 800–647–5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

For service information identified in this proposed rule, contact AgustaWestland, Product Support Engineering, Via del Gregge, 100, 21015 Lonate Pozzolo (VA) Italy, ATTN: Maurizio D’Angelo; telephone 39–0331–664757; fax 39 0331–664680; or at <http://www.agustawestland.com/technical-bulletins>. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Parkway, Fort Worth, Texas 76177.

FOR FURTHER INFORMATION CONTACT: Matt Wilbanks, Aviation Safety Engineer, Safety Management Group, Rotorcraft Directorate, FAA, 10101 Hillwood Parkway, Fort Worth, Texas 76177; telephone (817) 222–5110; email matt.wilbanks@faa.gov.

SUPPLEMENTARY INFORMATION:**Comments Invited**

We invite you to participate in this rulemaking by submitting written comments, data, or views. We also invite comments relating to the economic, environmental, energy, or federalism impacts that might result from adopting the proposals in this document. The most helpful comments reference a specific portion of the proposal, explain the reason for any recommended change, and include supporting data. To ensure the docket does not contain duplicate comments, commenters should send only one copy of written comments, or if comments are filed electronically, commenters should submit only one time.

We will file in the docket all comments that we receive, as well as a report summarizing each substantive public contact with FAA personnel concerning this proposed rulemaking. Before acting on this proposal, we will consider all comments we receive on or before the closing date for comments. We will consider comments filed after the comment period has closed if it is possible to do so without incurring expense or delay. We may change this proposal in light of the comments we receive.

Discussion

EASA, which is the Technical Agent for the Member States of the European Union, has issued EASA AD No. 2011-0207, dated October 20, 2011 (EASA 2011-0207), to correct an unsafe condition for certain serial-numbered Agusta Model AB139 and AW139 helicopters. An accident involving a Model AW139 helicopter caused the tail rotor (T/R), the T/R gearbox, and part of the fin to detach from the aircraft, rupturing the hydraulic lines and draining all of the hydraulic fluid. Investigation following the accident resulted in an assessment of the helicopter's hydraulic systems. According to EASA, this assessment revealed that an operational check of the hydraulic systems is necessary to ensure its functionality. EASA advises that this condition, if not corrected, could lead, in the case of multiple failures, to loss of hydraulic power and subsequent loss of control of the helicopter. To address this, EASA AD 2011-0207 requires, within 50 flight hours or 2 months, operational checks of the power control modules and shutoff valves and reporting the results to the manufacturer.

FAA's Determination

These helicopters have been approved by the aviation authority of Italy and are approved for operation in the United States. Pursuant to our bilateral agreement with Italy, EASA, its technical representative, has notified us of the unsafe condition described in its AD. We are proposing this AD because we evaluated all known relevant information and determined that an unsafe condition is likely to exist or develop on other products of the same type design.

Related Service Information Under 1 CFR Part 51

We reviewed Agusta Bollettino Tecnico No. 139-269, dated September 30, 2011 (BT 139-269), for Model AB139 and AW139 helicopters. BT 139-269 contains procedures for conducting operational checks of both hydraulic systems to confirm correct functionality. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Proposed AD Requirements

This proposed AD would require, within 50 hours time-in-service (TIS), performing operational tests of the Number 1 and Number 2 hydraulic systems power control modules (PCMs), the tail shut-off valve, the PCM1 and PCM2 flight control shut-off valves, and the emergency landing gear shut-off valve for correct functionality. Depending on the results of the the operational checks, this proposed AD would require replacing a PCM, the tail shut-off valve, a flight control shut-off valve, the number 2 hydraulic control panel, the number 1 hydraulic module, the number 1 or number 2 PCM pressure switch, or repairing the electrical wiring.

Differences Between This Proposed AD and the EASA AD

The EASA AD requires reporting the results of the operational checks to Agusta, while the proposed AD does not. The EASA AD also requires compliance within 50 flight-hours or 2 months, while the proposed AD requires compliance within 50 hours TIS.

Costs of Compliance

We estimate that this proposed AD would affect 102 helicopters of U.S. Registry. Based on an average labor rate of \$85 per hour, we estimate that operators may incur the following costs in order to comply with this proposed AD.

Performing the operational checks of the hydraulic systems would require about 2 work-hours for a total cost per helicopter of \$170 and a total cost to U.S. operators of \$17,340.

Replacing a PCM would require about 3 work-hours and required parts would cost about \$87,136, for a cost per helicopter of \$87,391.

Replacing a tail or flight control shut-off valve would require about 2 work-hours, and required parts would cost about \$7,512, for a cost per helicopter of \$7,682.

Replacing the number 2 hydraulic control panel would require about 2 work-hours, and required parts would cost about \$8,165, for a cost per helicopter of \$8,335.

Replacing the number 1 hydraulic module would require about 4 work-hours, and required parts would cost about \$87,137, for a cost per helicopter of \$87,477.

Replacing a PCM pressure switch would require about 2 work-hours, and required parts would cost about \$6,974, for a cost per helicopter of \$7,144.

Repairing the electrical wiring would require about 2 work-hours, and required parts would cost about \$45, for a cost per helicopter of \$215.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this proposed AD would not have federalism implications under Executive Order 13132. This proposed AD would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed, I certify this proposed regulation:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this proposed AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

The Proposed Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA proposes to amend 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

Agustawestland S.P.A. (Agusta): Docket No. FAA-2016-4278; Directorate Identifier 2012-SW-022-AD.

(a) Applicability

This AD applies to Agusta Model AB139 and AW139 helicopters, all serial numbers except serial number 31007, 31094, 31293, 31301, 31303, 31313, and 31329, certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as an inoperative hydraulic shut-off valve, which could result in loss of hydraulic power and subsequent loss of control of the helicopter.

(c) Comments Due Date

We must receive comments by May 10, 2016.

(d) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(e) Required Actions

Within 50 hours time-in service:

- (1) Perform an operational test of each Number 1 and Number 2 power control

module (PCM). If the fluid level in the reservoir changes more than 5mm (0.196 in) in an hour, replace the affected PCM.

(2) Perform an operational test of each tail shut-off valve. If the 2 SERVO caution message is not illuminated and the UTIL SOV2 and TR SOV indications are in the open position:

(i) Disconnect the Tail Shutoff valve connector, HP4P1.

(ii) Disconnect the PCM2 connectors, A44P3 and A44P12.

(iii) Disconnect the TB38 terminal board connector, TB38P1.

(iv) Perform a continuity test from HP4P1-1 to A44P12-16, from HP4P1-2 to TB38P1-D, and from HP4P1-4 to A44P3-6.

(v) If there is no continuity, repair or replace the defective wiring.

(vi) If there is continuity, release the test lever of the PCM2 to the DOWN NORM position.

(vii) If the TRSVO indication stays in the closed position, replace the tail shutoff valve.

(3) Perform an operational test of the PCM 2 flight control shut-off valve as described in the Compliance Instructions, paragraphs 5.1. through 5.5., of Agusta Bollettino Tecnico No. 139-269, dated September 30, 2011 (BT 139-269).

(i) If the 2 SERVO caution message is illuminated:

(A) On the hydraulic control panel, lift the guard of the SOV1/SOV2 switch and set it to SOV2 (closed position). Make sure that the 2 HYD PRESS caution message and the HYD 2 PRESS warning light on the hydraulic control panel are illuminated.

(B) Reset the SOV1/SOV2 switch to the open position.

(C) If the 2 HYD PRESS and 2 SERVO caution messages remain illuminated:

(1) Disconnect the PL14P1 and PL14P2 connectors from the hydraulic control panel.

(2) Disconnect the A1-1P4 connector from the MAU1.

(3) Disconnect the A2-1P3 connector from the MAU2.

(4) Disconnect the A44P3 connector from the Number 2 PCM.

(5) Disconnect the PL1P3 connector from the circuit breaker panel.

(6) Perform a continuity test from PL14P1-J to A1-1P4-18, from PL14P1-D to PL1P3-q, from PL14P2-J to A44P3-5, and from PL14P2-T to A2-1P3-34. If there is no continuity, repair or replace the defective wiring.

(7) If the HYD PRESS and 2 SERVO caution messages remain illuminated, replace the number 2 hydraulic power module.

(ii) If the 2 HYD PRESS caution message is illuminated, the HYD 2 pressure indication is more than 190 bar (2,755 lbf/sq in), and the SOV2 shutoff valve is in the open position, replace the pressure switch on the Number 2 PCM.

(iii) If the closure of SOV 2 is indicated on the MFD hydraulic synoptic page, before further flight, replace the Number 2 PCM.

(4) Perform an operational test of the PCM 1 flight control shut-off valve as described in the Compliance Instructions, paragraphs 6.1. through 6.4., of BT 139-269.

(i) If the 1 SERVO caution message is illuminated:

(A) On the hydraulic control panel, lift the guard of the SOV1/SOV2 switch and set it to SOV1 (closed position). Make sure that the 1 HYD PRESS caution message and the HYD 1 PRESS warning light on the hydraulic control panel are illuminated.

(B) Reset the SOV1/SOV2 switch to the open position. If the 1 HYD PRESS and 1 SERVO caution messages remain illuminated:

(1) Disconnect the PL14P1 and PL14P2 connectors from the hydraulic control panel.

(2) Disconnect the A1-1P4 connector from the MAU1.

(3) Disconnect the A2-1P3 connector from the MAU2.

(4) Disconnect the A45P3 connector from the Number 1 PCM.

(5) Disconnect the PL1P3 connector from the circuit breaker panel.

(6) Perform a continuity test from PL14P1-J to A1-1P4-18, from PL14P1-E to A45P3-5, from PL14P1-D to PL1P3-q, and from PL14P2-T to A2-1P3-34. If there is no continuity, repair or replace the defective wiring.

(7) If the HYD PRESS and 1 SERVO caution messages remain illuminated, replace the Number 1 hydraulic control panel.

(ii) If the 1 HYD PRESS caution message is illuminated, the HYD 1 pressure indication is more than 190 bar (2,755 lbf/sq in), and the SOV1 shutoff valve is in the open position, replace the pressure switch on the Number 1 PCM.

(iii) If the closure of SOV 1 is indicated on the MFD hydraulic synoptic page, before further flight, replace the Number 1 PCM.

(4) Perform an operational test of the emergency landing gear shutoff valve as described in the Compliance Instructions, paragraphs 7.1. through 7.4., of BT 139-269.

(i) If the EMERG L/G PRESS caution message is illuminated, the HYD 1 pressure indication is more than 190 bar (2,755 lbf/sq in), and the UTIL SOV1 (LDG GEAR EMER) shutoff valve is in the open position, replace the pressure switch on the Number 1 PCM.

(ii) If the 1 HYD MIN caution message is illuminated, inspect the fluid level on the Number 1 PCM and inspect the Number 1 main hydraulic system for leaks.

(A) If the fluid level is between the FULL and ADD marks, or if there are no hydraulic fluid leaks, perform an operational test of the level switches. If the 1 HYD MIN caution message is illuminated, replace the Number 1 PCM.

(B) If there is a hydraulic fluid leak:

(1) Replace all leaking parts and lines or repair the leak.

(2) If the 1 HYD MIN caution message remains illuminated, perform an operational test of the level switches.

(3) If the 1 HYD MIN caution message remains illuminated, replace the Number 1 PCM.

(f) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Group, FAA, may approve AMOCs for this AD. Send your proposal to Matt Wilbanks, Aviation Safety Engineer, Safety Management Group, Rotorcraft Directorate, FAA, 10101 Hillwood Parkway, Fort Worth, Texas 76177;

telephone (817) 222-5110; email 9-ASW-FTW-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office before operating any aircraft complying with this AD through an AMOC.

(g) Additional Information

The subject of this AD is addressed in European Aviation Safety Agency (EASA) AD No. 2011-0207, dated October 20, 2011. You may view the EASA AD on the Internet at <http://www.regulations.gov> in Docket No. FAA-2016-4278.

(h) Subject

Joint Aircraft Service Component (JASC) Code: 2900: Hydraulic Power.

Issued in Fort Worth, Texas, on March 3, 2016.

Scott A. Horn,

Acting Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 2016-05368 Filed 3-10-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2006-25513; Directorate Identifier 99-NE-61-AD]

RIN 2120-AA64

Airworthiness Directives; Rolls-Royce Deutschland Ltd & Co KG Turbofan Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: We propose to supersede airworthiness directive (AD) 2006-18-14 that applies to all Rolls-Royce Deutschland Ltd & Co KG (RRD) Tay 650-15 and Tay 651-54 turbofan engines. AD 2006-18-14 requires calculating and re-establishing the cyclic life of stage 1 high-pressure turbine (HPT) disks, part number (P/N) JR32013 and P/N JR33838, and stage 1 low-pressure turbine (LPT) disk, P/N JR32318A. This proposed AD would require re-calculating the cyclic life, and would impose a reduced cyclic life, of stage 1 HPT disk, P/N JR32013. We are proposing this AD to prevent failure of stage 1 HPT disks, P/N JR32013 and P/N JR33838, and stage 1 LPT disk, P/N JR32318A, which could result in an uncontained engine failure and damage to the airplane.

DATES: We must receive comments on this proposed AD by May 10, 2016.

ADDRESSES: You may send comments, using the procedures found in 14 CFR 11.43 and 11.45, by any of the following methods:

- *Federal eRulemaking Portal:* Go to <http://www.regulations.gov>. Follow the instructions for submitting comments.
- *Fax:* 202-493-2251.
- *Mail:* U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.
- *Hand Delivery:* Deliver to Mail address above between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays.

For service information identified in this proposed AD, contact Rolls-Royce Deutschland Ltd & Co KG, Eschenweg 11, Dahlewitz, 15827 Blankenfelde-Mahlow, Germany; phone: 49-0-33-7086-1064; fax: 49-0-33-7086-3276. You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2006-25513; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this proposed AD, the mandatory continuing airworthiness information, regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is in the **ADDRESSES** section. Comments will be available in the AD docket shortly after receipt.

FOR FURTHER INFORMATION CONTACT: Philip Haberlen, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7770; fax: 781-238-7199; email: philip.haberlen@faa.gov.

SUPPLEMENTARY INFORMATION:

Comments Invited

We invite you to send any written relevant data, views, or arguments about this proposed AD. Send your comments to an address listed under the **ADDRESSES** section. Include "Docket No. FAA-2006-25513; Directorate Identifier 99-NE-61-AD" at the beginning of your comments. We specifically invite

comments on the overall regulatory, economic, environmental, and energy aspects of this proposed AD. We will consider all comments received by the closing date and may amend this proposed AD because of those comments.

We will post all comments we receive, without change, to <http://www.regulations.gov>, including any personal information you provide. We will also post a report summarizing each substantive verbal contact we receive about this proposed AD.

Discussion

On August 30, 2006, we issued AD 2006-18-14, Amendment 39-14753 (71 FR 52988, September 8, 2006), for all RRD Tay 650-15 and Tay 651-54 turbofan engines. AD 2006-18-14 requires calculating and re-establishing the cyclic life of stage 1 HPT disks, P/N JR32013 and P/N JR33838, and stage 1 LPT disk, P/N JR32318A, that have been exposed to different engine flight plan profiles. AD 2006-18-14 also requires removing from service, using a drawdown schedule, those stage 1 HPT disks and stage 1 LPT disks operated under Tay 650-15 engine flight plan profiles A, B, C, or D; or operated under the Tay 651-54 engine datum flight profile, at reduced cyclic life limits. AD 2006-18-14 resulted from RRD updating their low-cycle-fatigue analysis for stage 1 HPT disks and stage 1 LPT disks and reducing their cyclic life limits. We issued AD 2006-18-14 to prevent cracks leading to turbine disk failure, which could result in an uncontained engine failure and damage to the airplane.

Actions Since AD 2006-18-14 Was Issued

Since we issued AD 2006-18-14, RRD reviewed the cyclic life limit of parts affected by AD 2006-18-14; RRD concluded that the stage 1 HPT disk, P/N JR32013, requires further cyclic life limit reduction. RRD did not further reduce the cyclic life limit of stage 1 HPT disk, P/N JR33838, or stage 1 LPT disk, P/N JR32318A. Accordingly, the cyclic life limits of stage 1 HPT disk, P/N JR33838, and stage 1 LPT disk, P/N JR32318A, as imposed by AD 2006-18-14, remain unchanged in this proposed AD.

Since AD 2006-18-14 was issued, the European Aviation Safety Agency (EASA) issued AD 2015-0056, dated March 31, 2015 to reduce the cyclic life limits of the stage 1 HPT disk, P/N JR32013.